

**REMARKS/ARGUMENTS**

The Office Action mailed September 3, 2003, has been received and reviewed. Claims 1 through 20 are currently pending in the application, of which claims 1, 2, 3, 6, and 7 are currently under examination. Claims 4, 5, and 8 through 20 were previously withdrawn from consideration as being drawn to non-elected inventions and have been cancelled herein without prejudice or disclaimer. Each of claims 1, 2, 3, 6, and 7 stands rejected. Claims 1, 2, 3, 6, and 7 have been amended herein and claim 21 has been added. Applicants respectfully request reconsideration of the application in light of the above amendments and the following remarks.

**Information Disclosure Statement(s)**

Applicants note the filing of an Information Disclosure Statement herein on July 30, 2001, and note that although a copy of the PTO/SB/08 was returned with the outstanding Office Action, a foreign patent document (FR 2554114) was not considered by the Examiner because no translation was provided. Applicants apologize for this omission, and resubmit the reference with an English translation in conjunction with a Supplemental Information Disclosure Statement. It is respectfully requested that FR 2554114 be considered and made of record herein and that an initialed copy of the PTO/SB/08 evidencing the same be returned to the undersigned attorney.

**Amendments to the Title**

The title has been amended herein to correct a minor editorial error. It is respectfully submitted that no new matter has been added by way of the title amendment.

**Amendments to the Specification**

Amendments to the specification have been made to correct minor editorial problems. It is respectfully submitted that the amendments are supported by the as-filed specification and no new matter has been added.

**35 U.S.C. § 102(b) Anticipation Rejections****A. Applicable Authority**

A claim is anticipated only if each and every element as set forth in the claim is found, either expressly or inherently described, in a single prior art reference. *Verdegaal Brothers v. Union Oil Co. of California*, 2 USPQ2d 1051, 1053 (Fed. Cir. 1987). The identical invention must be shown in as complete detail as is contained in the claim. *Richardson v. Suzuki Motor Co.*, 9 USPQ2d 1913, 1920 (Fed. Cir. 1989).

**B. Anticipation Rejection Based on U.S. Patent 5,840,227 to Bourdoncle et al.**

Claim 1 stands rejected under 35 U.S.C. § 102(b) as being anticipated by U.S. Patent 5,840,227 to Bourdoncle et al. (hereinafter the “Bourdoncle reference”). As the Bourdoncle reference fails to describe, either expressly or inherently, each and every element as set forth in claim 1 of the above-referenced application, Applicants respectfully traverse this rejection, as hereinafter set forth.

As stated in the Specification, it is known in the art to use elastomeric compositions in the manufacture of rocket motor insulation. *See, Substitute Specification at ¶ [0008]*. However, the ablative properties of elastomers are often inadequate for rocket motor application. *See id.* As such, elastomeric compositions are sometimes reinforced with fibers, such as carbon fibers, to improve the ablative properties thereof. *See id. at ¶ [0009]*. Due to the fragility of carbon fibers, however, conventional methods for achieving homogeneous dispersion of carbon fibers in an elastomeric composition often deteriorate the carbon fibers into smaller particles or shreds, thereby negating the advantageous physical attributes that the carbon fibers would otherwise have contributed to the insulation. *See, Substitute Specification at ¶ [0011]*. The inventors hereof have recognized that if a liquid elastomeric composition is used in the manufacture of rocket motor insulation such that it comprises at least 50% of the elastomer present in the insulation, the carbon fibers may be introduced under substantially solvent-free conditions without substantial fragmentation of the fibers. *See id. at ¶ [0017]*. Accordingly, the present invention provides a method for insulating a case of a solid propellant rocket motor wherein the insulation is prepared from a cross-linkable liquid polymer, e.g., liquid ethylene propylene diene monomer (EPDM),

and carbon fibers under substantially solvent-free conditions (*i.e.*, not more than about 5 weight percent of volatile solvent based on the dry ingredients of the insulation). *See id.* at ¶ [0030]. Once the insulation has been prepared according to the methods of the present invention, it may be shaped into sheets and either stored or molded onto an interior surface of the rocket motor case, between the rocket motor case and the propellant. *See id.* at ¶¶ [0056] and [0057]; FIGs. 1A and 1B.

It is respectfully submitted that the prior art references cited in the outstanding Office Action fail to disclose a method of insulating a rocket motor case with sheets of insulation manufactured such that at least 50% of the elastomer, *e.g.*, EPDM terpolymer, in the insulation is provided in liquid form, and wherein carbon fibers are introduced into the elastomer under substantially solvent-free conditions.

Independent claim 1, as amended herein, recites a method of insulating a rocket motor case loaded with a solid propellant. The method comprises preparing insulation from a composition comprising a cross-linkable liquid polymer and carbon fibers, curing the composition to form the insulation, shaping the insulation into at least one sheet, and insulating the rocket motor with the at least one sheet of insulation. The composition from which the insulation is prepared includes the cross-linkable liquid polymer in a sufficient concentration to permit the carbon fibers to be dispersed therein by mixing under substantially solvent-free conditions. It is made clear from the Specification of the present application, that a concentration of cross-linkable liquid polymer sufficient to permit the carbon fibers to be dispersed therein under substantially solvent-free conditions is at least about 50% by weight of the total cross-linkable polymer used (*i.e.*, solid and liquid cross-linkable polymer), and preferably at least about 90% by weight. *See, Substitute Specification* at ¶ [0031].

The Bourdoncle reference, on the other hand, discloses a method of making internal thermal protection for a thruster structure. *See, Bourdoncle reference* at col. 2, lines 44–49. The method comprises making an elastomer base and mixing it with fibers, *e.g.*, carbon fibers, in order to obtain thermal protection in the form of a fiber-reinforced elastomer layer. *See id.* The thermal protection is subsequently molded *in situ* on the structure to be protected, *i.e.*, the thruster structure. *See id.* at col. 2, lines 58–60. More particularly, the thermal protection is

molded on the inside wall of an end wall portion 11 and an extender portion 12 of a metal end wall and extender structure 10 which is disposed between a thruster body 18 housing the propellant and a nozzle 19 for ejecting the combustion gases produced by the propellant. *See id.* at col. 3, lines 49–65.

The elastomer base of the internal thermal protection has a viscosity of less than 1500 Pa. at 50°C and is essentially constituted by a substance selected from polychloroprenes and silicone gums that vulcanize at a temperature above 120°C. *See id.* at col. 3, lines 50–54. While the low viscosity of the elastomer base makes it possible to perform the mixing of the carbon fibers by means of a mechanical mixer without putting the elastomer base into solution, it also renders the insulation unsuitable for providing sheets that can be molded onto a surface. *See id.* at col. 2, line 63–col. 3, line 5.

It is respectfully submitted that the Bourdoncle reference does not disclose, either expressly or inherently, a method of insulating a rocket motor case loaded with a solid propellant, which comprises, in part, *shaping the insulation into at least one sheet* and insulating the rocket motor case with the at least one sheet of insulation, as recited in amended independent claim 1 of the present application. Rather, the Bourdoncle reference specifically states that insulation manufactured from the low viscosity elastomer bases disclosed therein are “unsuitable for providing sheets” that can be molded onto a surface. *Id.*

Accordingly, it is respectfully submitted that the Bourdoncle reference does not disclose, either expressly or inherently, each and every element as set forth in independent claim 1, as amended herein and, as such, the Bourdoncle reference does not anticipate this claim. Accordingly, Applicants respectfully request withdrawal of the 35 U.S.C. § 102(b) rejection of independent claim 1. Claim 1 is believed to be in condition for allowance and such favorable action is respectfully requested.

### **35 U.S.C. § 103(a) Obviousness Rejections**

#### **A. Applicable Authority**

The basic requirements of a *prima facie* case of obviousness are summarized in MPEP §2143 through §2143.03. In order “[t]o establish a *prima facie* case of obviousness, three basic

criteria must be met. First, there must be some suggestion or motivation, either in the references themselves or in the knowledge generally available to one of ordinary skill in the art, to modify the reference or combine the reference teachings. Second, there must be a reasonable expectation of success [in combining the references]. Finally, the prior art reference (or references when combined) must teach or suggest all the claim limitations. The teaching or suggestion to make the claimed combination and the expectation of success must both be found in the prior art, not in applicant's disclosure. *In re Vaeck*, 947 F.2d 488, 20 USPQ2d 1438 (Fed. Cir. 1991)." Further, in establishing a *prima facie* case of obviousness, the initial burden is placed on the Examiner. "To support the conclusion that the claimed invention is directed to obvious subject matter, either the references must expressly or impliedly suggest the claimed invention or the examiner must present a convincing line of reasoning as to why the artisan would have found the claimed invention to have been obvious in light of the teachings of the references." *Ex parte Clapp*, 227 USPQ 972, 973 (Bd. Pat. App. & Inter. 1985). See also MPEP §706.02(j) and §2142.

B. Obviousness Rejection Based on U.S. Patent 5,840,227 to Bourdoncle et al. in View of the Admitted Prior Art or U.S. Patent 6,606,852 to Harvey et al.

Claims 2, 3, 6, and 7 stand rejected under 35 U.S.C. § 103(a) as being unpatentable over the Bourdoncle reference in view of the admitted prior art as set forth at page 8, lines 23–24 of the present application (hereinafter the "APA"), or U.S. Patent 6,606,852 to Harvey et al. (hereinafter the "Harvey reference"). As a *prima facie* case of obviousness cannot be established based upon the asserted combination of references, Applicants respectfully traverse this rejection, as hereinafter set forth.

Independent claim 2, as amended herein, recites a method of insulating a rocket motor case loaded with a solid propellant. The method comprises preparing insulation from a composition comprising a cross-linkable EPDM terpolymer and carbon fibers, curing the composition to form the insulation, and insulating the rocket motor case with the insulation. The cross-linkable EPDM terpolymer from which the insulation is prepared comprises a

cross-linkable liquid EPDM terpolymer in a sufficient concentration to permit the carbon fibers to be dispersed into the composition by mixing under substantially solvent-free conditions.

The Bourdoncle reference was discussed hereinabove. In contrast to independent claim 2, the Bourdoncle reference fails to teach or suggest a method of insulating a rocket motor case loaded with a solid propellant which comprises, in part, preparing insulation from a composition comprising a cross-linkable EPDM terpolymer, as recited in amended independent claim 2. Rather, the elastomer base of the thermal protection disclosed in the Bourdoncle reference is essentially constituted by a substance selected from polychloroprenes and silicone gums. *See id.* at col. 3, lines 50–54.

It is stated in the outstanding Office Action that while the Bourdoncle reference does not teach using a cross-linkable EPDM terpolymer as the cross-linkable elastomer base, it would have been obvious to one of ordinary skill in the art at the time the invention was made to use the cross-linkable liquid EPDM terpolymer of either the APA or the Harvey reference as the cross-linkable polymer of the Bourdoncle reference “in order to produce a durable and effective insulation layer.” *Office Action* at page 3, ¶ 7. Applicants respectfully traverse this assertion, as hereinafter set forth.

At page 8, lines 23–24 of the as-filed application, it is stated that TRILENE® 67A and TRILENE® 77 are liquid EPDM terpolymers that are commercially available through Uniroyal Chemical Company of Middlebury, Connecticut. *See, Substitute Specification* at ¶ [0032]. However, conventional uses of the disclosed liquid EPDM terpolymers with regard to rocket motor insulation are not disclosed.

The Harvey reference discloses an elastomer-based insulation for rocket motors which may comprise, in part, liquid EPDM (TRILENE® 67A). *See, Harvey reference* at col. 1, lines 13–14; col. 8, lines 24–50 (Table 1). The liquid EPDM of the Harvey reference comprises between 15% and 30% of the total EPDM utilized in preparation of the insulation. *See id.* at Table 1. The remainder of the EPDM utilized is in solid form (*i.e.*, NORDEL® 1040 and NORDEL® 2522). *See id.*

It is respectfully submitted that a *prima facie* case of obviousness cannot be established for amended independent claim 2 based upon the combination of the Bourdoncle reference and

the APA or the Harvey reference as the asserted combination of references fails to teach or suggest all of the limitations of this claim. The Bourdoncle reference teaches the manufacture of insulation for a thruster structure wherein essentially 100% of the insulation is essentially constituted by a substance selected from polychloroprenes and silicon gums. *See, Bourdoncle reference* at col. 3, lines 50-54. The Harvey reference teaches the use of liquid EPDM in the manufacture of rocket motor insulation in a concentration between 15% and 30% and in a combination with a relatively higher concentration of solid EPDM. However, none of the art cited, nor the combination thereof, teaches or suggests the use of liquid EPDM in the manufacture of rocket motor insulation in a concentration sufficient to permit the carbon fibers to be dispersed into the composition by mixing under substantially solvent free conditions (*i.e.*, a concentration of at least 50%), as recited in independent claim 2, as amended herein.

Additionally, it is respectfully submitted that there is no suggestion or motivation, either in the references themselves or in the knowledge generally available to one of ordinary skill in the art, to combine the reference teachings as asserted. *See, In re Vaeck*, 947 F.2d 488, 20 UPSQ2d 1438 (Fed. Cir. 1991). The Bourdoncle reference merely teaches that utilizing a low viscosity polychloroprene or silicone gum in the manufacture of insulation for a thruster structure permits fibers to be mixed therewith by mechanical means without resorting to dissolving the elastomer base and without damaging the fibers. *See, Bourdoncle reference* at col. 4, lines 14-25. It does not suggest that this observation would hold true if liquid EPDM were substituted for the low viscosity polychloroprene or silicone gum.

Further, the Harvey reference teaches only that liquid EPDM is suitable for manufacturing rocket motor insulation if present in a low concentration (*i.e.*, a concentration between 15% and 30%, based on the total EPDM utilized) and in combination with a relatively larger concentration of solid EPDM. *See, Harvey reference* at Table 1. It does not suggest that a greater concentration of liquid EPDM would be suitable nor does it suggest that liquid EPDM in the substantial absence of solid EPDM would be suitable.

In view of the foregoing, Applicants respectfully submit that the cited references fail to establish a *prima facie* case of obviousness of independent claim 2, as amended herein. Accordingly, it is respectfully requested that the rejection under 35 U.S.C. § 103(a) of claim 2

based upon the asserted combination of references be withdrawn. Claim 2 is believed to be in condition for allowance and such favorable action is respectfully requested.

Each of dependent claims 3, 6, and 7 is also believed to be in condition for allowance for at least the above-cited reasons as the independent claim from which they depend, either directly or indirectly, is in condition for allowance. *See, In re Fine*, 837 F.2d 1071, 5 USPQ2d 1596, 1600 (Fed. Cir. 1988) (dependent claims are nonobvious under 35 U.S.C. § 103 if the independent claims from which they depend are nonobvious). Thus, it is respectfully requested that the obviousness rejections of claims 3, 6, and 7 be withdrawn as well. Each of claims 3, 6, and 7 is believed to be in condition for allowance and such favorable action is respectfully requested.

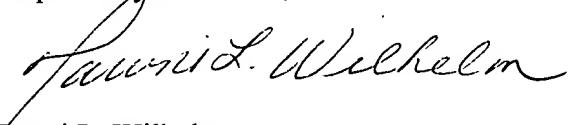
**NEW CLAIMS**

New dependent claim 21 has been added herein. It is respectfully submitted that claim 21 is supported by the as-filed specification and no new matter has been added. Claim 21 is believed to be in condition for allowance and such favorable action is respectfully requested.

**CONCLUSION**

Each of claims 1, 2, 3, 6, 7, and 21 is believed to be in condition for allowance, and an early notice thereof is respectfully solicited. Should it be determined that additional issues remain which might be resolved by a telephone conference, the Examiner is respectfully invited to contact Applicants' undersigned attorney.

Respectfully submitted,



Tawni L. Wilhelm  
Registration Number 47,456  
Attorney for Applicant  
TRASKBRITT, PC  
P.O. Box 2550  
Salt Lake City, Utah 84110  
Telephone: (801) 532-1922

TLW/ps:rmh:tlb

Date: December 3, 2003

Attachments: Appendix A – Clean Version of Substitute Specification  
Appendix B – Redlined Version of Substitute Specification  
Appendix C – Replacement Sheets and Annotated Sheets Showing Changes  
(drawings)

Document in ProLaw

# **APPENDIX B**

**(VERSION OF SUBSTITUTE SPECIFICATION  
WITH MARKINGS TO SHOW CHANGES MADE)**

**(Serial No. 09/747,192)**

PATENT  
Attorney Docket 2507-5730.1US (21761-US-01)

NOTICE OF EXPRESS MAILING

Express Mail Mailing Label Number: \_\_\_\_\_

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APPLICATION FOR LETTERS PATENT

for

**METHOD OF INSULATING CASE A CASE OF OF A SOLID  
PROPELLANT ROCKET MOTOR**

Inventor:  
David G. Guillot

Attorney:  
Joseph A. Walkowski  
Registration No. 28,765  
**TRASKBRITT**  
P.O. Box 2550  
Salt Lake City, Utah 84110  
(801) 532-1922

METHOD OF INSULATING ~~CASE A CASE OF OF~~ A SOLID  
PROPELLANT ROCKET MOTOR

CROSS REFERENCE TO RELATED APPLICATION

[0001] Priority is claimed of U.S. provisional application 60/171,619 filed in the U.S. Patent & Trademark Office on December 23, 1999, the complete disclosure of which is incorporated herein by reference.

ORIGIN OF THE INVENTION STATEMENT REGARDING FEDERALLY SPONSORED  
RESEARCH OR DEVELOPMENT

[0002] The U.S. Government has a paid-up license in this invention and the right in limited circumstances to require the patent owner to license others on reasonable terms as provided by the terms of F04611-97-C-0053 to the Air Force Rocket Laboratory.

BACKGROUND OF THE INVENTION

1. Field of the Invention

[0003] Field of the Invention: This invention is directed to a process for making elastomer-based insulation for rocket ~~motors, motors and and, in particular~~ particular, to a process in which fragile carbon fibers are mixed with and preferably homogeneously dispersed in ethylene propylene diene monomer-EPDM (EPDM) without requiring the use of a volatile solvent for dissolution of EPDM during fiber incorporation. The insulation of this invention is especially useful for placement in the nozzle or case, including between a solid propellant grain and a rocket motor case for protecting the case from high temperatures experienced during burning of solid propellant grains.

2. Description of the Related Art

[0004] State of the Art: Solid rocket motors typically include an outer case or shell housing a solid propellant grain. The rocket motor case is conventionally manufactured from a rigid, yet durable, material such as steel or filament-wound composite. The propellant is housed within the case and is formulated from a composition designed to undergo combustion and thereby produce the requisite thrust for attaining rocket motor propulsion.

**[0005]** During operation, a heat insulating layer (insulation) protects the rocket motor case from heat and ~~particles~~ particle streams generated by the combusting propellant. Typically, the insulation is bonded to the inner surface of the case and is generally fabricated from a composition capable of withstanding the high temperature gases produced when the propellant grain burns. A liner layer (liner) functions to bond the propellant grain to the insulating layer and to any ~~non-~~noninsulated portions of the case, as well as to inhibit interfacial burning. Liner compositions are generally known to those skilled in the art. An exemplary liner composition and process of applying the same to a case is disclosed in U.S. Patent №-5,767,221, the complete disclosure of which is incorporated herein by reference to the extent that it is compatible with this specification.

**[0006]** The combustion of solid rocket propellant generates extreme conditions within the case of the rocket motor. For example, temperatures inside the rocket motor case typically reach ~~2,760~~ 2,760°C (5,000°F), and interior pressures may exceed 1,500 psi. These factors combine to create a high degree of turbulence within the rocket motor case. In addition, particles are typically entrained in the gases produced during ~~the~~-propellant combustion. Under the turbulent environment, these entrained particles can erode the rocket motor insulation. If the insulating layer and liner are pierced during rocket motor operation, the casing ~~it is~~ susceptible to melting or degradation, which can result in failure of the rocket motor. Thus, it is crucial that insulation compositions withstand the extreme conditions experienced during propellant combustion and protect the case from the burning propellant. It is also crucial that insulation compositions possess acceptable shelf life characteristics such that ~~it~~ they ~~remain~~ remain sufficiently pliable, without becoming fully cured, until used in application to the rocket motor casing. This requirement is essential because the production of a given lot of insulation may have to wait in storage for a number of months prior to use. Typically, the insulation may be stored in large rolls in an uncured, or at most a partially cured, state until ready for use. A number of curing agents are well known and are conventionally employed but still must be compatible with the overall EPDM formulation to permit satisfactory shelf life. This in turn requires a balancing of curing agent activity.

**[0007]** In the past, attempts at producing insulating materials that would protect the rocket motor case focused on filled and unfilled rubbers and ~~plastics~~ plastics, such as, as

phenolic resins, epoxy resins, high temperature melamine-formaldehyde coatings, ceramics, polyester resins, and the like. These plastics, however, crack and/or blister as a result of the rapid temperature and pressure fluctuations experienced during combustion.

**[0008]** Elastomeric compositions have also been used as rocket motor insulation materials in a large number of rocket motors. The elastomeric compositions have been selected because their mechanical, thermal, and ablative properties are particularly suited for rocket motor applications. However, the ablative properties of elastomers are often inadequate for rocket motor operation. For example, insulation, whether thermosetting or thermoplastic, ~~are~~ is characterized by relatively high erosion rates unless reinforced with a suitable filler. The criticality of avoiding such high erosion rates is demonstrated by the severity and magnitude of the risk of failure due to erosion. Most insulation ~~is~~ is of ~~necessity~~ necessity, "man-rated" in the sense that a catastrophic failure can result in the loss of human life -- whether the rocket motor is used as a booster for launch of ~~the~~ a space shuttle or is carried tactically underneath the wing of an attack aircraft. The monetary cost of failure in satellite launches is well-publicized and can run into the hundreds of millions of dollars.

**[0009]** In order to improve the ablative properties of elastomeric compositions, it has been proposed to reinforce the elastomeric compositions with fillers, such as organic-based fibers or carbon fibers. For instance, an exemplary carbon fiber-filled rocket motor insulation composed of solid ~~NORDEL NORDEL®~~ 1040 as the primary terpolymer is commonly known in the industry as the STW4-2868 thermal insulation and has the following composition:

[0010] TABLE A

STW4-2868 THERMAL INSULATION FORMULATION  
(carbon fiber; parts by weight)

Ingredient	Function	Parts by Weight
<u>NORDEL</u> 1040	Primary EPDM terpolymer base	80
Neoprene FB	Secondary polymer base	20
Zinc oxide	Activator	5
Sulfur	Curative	1
HAF carbon black	Pigment	1
MBT	Accelerator	1
<u>AGERITE</u> ® Resin D	Antioxidant	2
<u>AGERITE</u> ® HPS	Antioxidant	1
Tellurac	Accelerator	0.50
Sulfads	Accelerator	0.75
VCM carbon fibers	Filler	41
Total Parts by Weight		153.25

[0011] Although organic-based fibers can be dispersed within the EPDM without too much difficulty, the homogeneous dispersion of carbon fibers in an elastomeric composition presents a difficult processing problem. The mixing process is complicated by the fragility of the carbon fibers. Mixing of carbon fibers into a solid elastomer under high shear physically deteriorates the carbon fibers into smaller particles or shreds, thereby negating the advantageous physical attributes that the carbon fibers would otherwise have contributed to the insulation.

[0012] Conventionally, the problem of carbon fiber fragility has been addressed by dissolving the elastomer ~~into~~ into a solution with an appropriate organic solvent to lower the viscosity of the elastomer or elastomer mixture. Suitable solvents include, by way of example, hydrocarbons such as hexanes, heptanes, and/or cyclohexane. The frangible graphitized carbon

fibers can then be mixed with the solution in, for example, a sigma-blade mixer without significant breakage of or damage to the carbon fibers. The material is then sheeted out and the solvent allowed to evaporate at ambient atmosphere or in an oven.

[0013] The use of solvent in this processing technique presents several drawbacks. For example, solvent processing techniques, such as those conventionally used to disperse carbon fibers in EPDM (~~ethylene propylene diene monomer~~) rubber, are relatively expensive. Material costs are increased by the use of solvents, as are processing costs, since additional workers and equipment are required to handle and process the solvents. Further, considerable costs and worker safety issues are associated with the disposal of hazardous volatile organic solvents.

[0014] Thus, although it has been long recognized that carbon fiber-filled EPDM is an excellent candidate for ~~such~~ rocket motor insulation, ~~a need persists in the art for a low cost and non-nonhazardous solvent-free synthesis route that produces EPDM insulation having carbon fibers homogeneously dispersed therein, but without being subject to significant breakage or damage~~ would be desirable.

## SUMMARY OF THE INVENTION

[0015] ~~It is, therefore, an object of this invention to fulfill the long felt need in the art outlined above by providing a method of manufacturing ethylene propylene diene monomer EPDM (EPDM) rocket motor insulation in which carbon fibers are dispersed and immobilized in the EPDM polymeric matrix, but are not excessively fractured or fragmentized, i.e., i.e., broken into smaller fragments, when encountering degrees of shear necessary to homogeneously or otherwise distribute or disperse the carbon fibers in the EPDM polymeric matrix is provided.~~

[0016] ~~The above and other objects are attained by~~ The method of the present invention is a substantially solvent-free process method in which the insulation is manufactured via distributive/reduced shear mixing to distribute the fragile carbon fibers into a rubber matrix without excessive damage.

[0017] In accordance with one embodiment of this substantially solvent-free process method, the elastomer composition comprises carbon fibers and EPDM terpolymer, at least 50 wt% of which is introduced as an ingredient into the mixing apparatus as liquid EPDM terpolymer having a sufficiently low molecular weight and high diene content to permit

dispersion of the carbon fibers in the EPDM without substantial fragmentation of the fibers. As referred to herein, ~~liquid EPDM~~ “liquid EPDM” means EPDM terpolymer that is flowable at room temperature. Suitable mixing apparatuses for this embodiment include sigma-blade and vertical-blade mixers. Certain kneaders, such as discussed below in connection with ~~the second another embodiment of the inventive method~~, capable of superimposing a rotational and axial mixing motion to the carbon fibers can also be used.

[0018] In accordance with another embodiment of the invention, the elastomer composition is prepared, optionally under substantially solvent-free conditions with little or no liquid EPDM terpolymer, by use of a kneader capable of rotating a screw having a discontinuous ~~screw-screw~~ thread about the screw axis while superimposing an axial axially reciprocating stroke to the screw. This kneader imparts low shear distributive mixing of the carbon fibers in the EPDM terpolymer. The kneader used in this embodiment is especially suitable where little or no liquid EPDM ingredient and no volatile solvent are included in the formulation.

[0019] As referred to herein, carbon fibers are fibers having been subject to at least substantial graphitization or carbonization, and preferably have about 98 wt% or more carbon content.

[0020] Other ~~objects~~-aspects and advantages of the invention will be more apparent to those skilled in the art upon reading the detailed description and appended claims which, when read in conjunction with the accompanying drawings, explain the principles of this invention.

#### BRIEF DESCRIPTION OF THE DRAWINGS

[0021] The accompanying drawings serve to elucidate the principles of this invention. In such drawings:

[0022] FIG.-1A is a schematic cross-sectional view of a rocket motor assembly in which the insulation is provided;

[0022A] FIG. 1B is an enlarged schematic cross-sectional view of the area encircled and labeled “SEE FIG. 1B” in FIG. 1A;

[0023] FIG. 2 is a schematic cross-sectional view of a kneading apparatus suited for use with this invention;

[0024] FIG. 3 is a cross-sectional view taken along sectional line III-III in FIG. 2;

[0025] FIG. 4 is a schematic view of an axial segment of a discontinuous screw barrel of the kneading apparatus of FIG. 2, with the axial segment section being projected onto a flat plane for explanatory purposes;

[0026] FIG. 5 is the schematic view of FIG. 4, with kneading pins of the kneading apparatus being superimposed onto the illustrated axial segment;

[0027] FIG. 6 is the schematic view of FIG. 5, showing the paths of relative movement of selected ones of the kneading pins relative to the discontinuous screw ~~barrel~~, ~~barrel-and-and~~, in ~~particular~~ particular, relative to the screw flights of the barrel; and

[0028] FIG. 7 is a schematic, cross-sectional view of a char motor used in testing examples discussed hereinbelow.

#### DETAILED DESCRIPTION OF THE INVENTION

[0029] The present insulation compositions 10, when in a cured state, are especially suited for disposal on the interior surface of the rocket motor case 12, as shown in ~~FIG.~~ FIGS. 1A and 1B. Typically, a liner 14 is interposed between the insulation 10 and propellant 16. The insulation 10 and the liner 14 serve to protect the case 12 from the extreme conditions produced by the propellant 16 as it undergoes combustion reactions and is exhausted through nozzle assembly ~~-20~~ 18. Methods for loading a rocket motor case 12 with the insulation 10, the liner 14, and the propellant 16 are known to those skilled in the art, and can be readily adapted within the skill of the art without undue experimentation to incorporate the insulation composition of this invention.

[0030] Unlike conventional techniques that make use ~~of~~ of a solvent within a mixing apparatus to achieve adequate distribution of carbon fibers in solid EPDM ingredients without significant fiber fragmentation, ~~it is an object~~ the method of this invention ~~to achieve~~ achieves distribution of carbon fibers in an EPDM matrix under solvent-free conditions, or at least substantially solvent-free conditions. As referred to herein, ~~substantially~~ "substantially solvent-free-free" means that the process is performed with a sufficiently small amount of volatile solvent that, even if the volatile solvent is not removed during manufacture of the insulation, the volatile solvent will not be present in a sufficient amount to violate applicable environmental or safety regulations during manufacture, rocket motor storage, or rocket motor

operation due to volatilizing of the solvent. Generally, the term substantially "substantially solvent-free" preferably means not more than about 5 wt% of volatile solvent based on the dry ingredients of the insulation. Preferably, the process is conducted completely free of volatile solvent.

[0031] In accordance with a first embodiment of this invention, this and other objects are achieved by using liquid EPDM as a significant portion of the EPDM ingredients introduced into the mixing apparatus. The amount of liquid EPDM ingredient used to ensure adequate distribution of the fibers, without accompanying excessive fragmentation of the fibers, depends upon the mixing apparatus used. Generally, where a conventional mixer known in the insulation industry, such as a sigma-blade mixer, is used to disperse the carbon fibers within the EPDM matrix, the insulation composition preferably contains at least about 50% by weight, and more preferably at least about 90% by weight liquid EPDM as an ingredient, based on the total weight of the EPDM (*i.e., i.e.*, both the solid and liquid EPDM ingredients). Where a vertical blade mixer is used to disperse the carbon fibers within the EPDM matrix, the insulation composition preferably contains slightly more liquid EPDM, such as at least about 90% by weight, and more preferably at least about 95% by weight liquid EPDM as an ingredient, based on the total weight of the EPDM (*i.e., i.e.*, both the solid and liquid EPDM ingredients). Where a kneader such as the one illustrated in FIGS. 2-6 is used, even less of the liquid EPDM (or even no liquid EPDM, as detailed in the second embodiment below) is required to obtain homogeneous dispersion of the fibers without excessive fragmentation, *i.e., i.e.*, all of the EPDM can be in a solid state when introduced into the kneader.

[0032] Generally, the EPDM, *i.e., i.e.*, both the solid and liquid ingredients, comprises from about 35 wt% to about 90 wt%, and still more preferably from about 45 wt% to about 75 wt% of the total weight of the rocket motor insulation. The EPDM terpolymer can be formed from 1,4-hexadiene, dicyclopentadiene, and/or an alkylidene norbornene, such as ethylidene norbornene (ENB), as the diene component. Suitable commercially available liquid EPDM terpolymers are TRILENE TRILENE® 67A and TRILENE TRILENE® 77, available through Uniroyal Chemical Company of Middlebury, Connecticut. It is noted, however, that a portion or all of the liquid EPDM can be substituted for another liquid polymer ingredient, such as liquid polyurethanes, so long as the substituted liquid polymer ingredient obtains the same

distributive function with regard to the carbon fibers without excessive fragmentation. Suitable solid EPDM terpolymers having a 1,4-hexadiene component for use in this invention include NORDEL NORDEL® 1040, NORDEL NORDEL® 2522, and NORDEL NORDEL® 2722E, made by DuPont Dow Elastomers of Wilmington, Delaware. Suitable solid EPDM terpolymers having an ENB diene component for use in this invention include, without limitation, and as stated above, KELTAN KELTAN® 4506, KELTAN KELTAN® 1446A, KELTAN KELTAN® 2308, each of which is available from DSM Elastomers of the Netherlands, and NORDEL NORDEL® IP 4520 and NORDEL NORDEL® IP 4640, both of which are and continue to be available from DuPont Dow Elastomers.

[0033] The curing package can include sulfur curing agents and/or peroxide curing agents for ~~cross~~cross-linking and/or chain extending polymers or polymer precursors (e.g., prepolymers). A ~~suitable~~Suitable insoluble sulfur curing agent ~~agents is are~~agents is ~~AKROSPERSE AKROSPERSE® IS-70 from Akro Chem, Akrochem Corporation of Akron, Ohio, and CRYSTEX CRYSTEX® OT-20 available through Charles H. Haynes, Inc.~~ Other forms of elemental sulfur can also be used. Suitable peroxide curing agents include dicumyl peroxide, 2,5-dimethyl-2,5-bis-(t-butylperoxy)hexane, 2,5-dimethyl-2,5-bis-(benzoylperoxy)hexane, 2,5-dimethyl-2,5-di(t-butylperoxy)-3-hexane, n-butyl-4,4-bis-(t-butylperoxyl)valerate, 4,4'-methyl-bis-(cyclohexylamine)carbomate, 1,1-bis-(t-butylperoxy)-3,3,5-trimethylcyclohexane,  $\alpha,\alpha'$ -bis-(t-butylperoxy)-diisopropylbenzene, 2,5-dimethyl-2,5-bis-(t-butylperoxy)hexyne-3, and t-butyl perbenzoate. A commercially available peroxide is available under the trade name Di-Cup DI-CUP® 40KE, which comprises about 40% dicumyl peroxide on a clay carrier. (The clay carrier is available from Burgess Pigment Company of Sandersville, Georgia.) Another suitable curing agent (besides sulfur and peroxide curing agents) is bromomethyl alkylated phenolic resin, available as SP-1056 from Schenectady Int'l, Inc. of Schenectady, N.Y.

[0034] In typical formulations, the curing agent comprises from about 0.5 phr to about 8 phr, more preferably about 2 phr to about 5 phr. As referred to herein and generally accepted in the art, "phr" means parts by weight per one hundred parts by weight polymer.

[0035] The curing package preferably also includes at least one phosphate cure accelerator. In the case of a sulfur curing agent, the accelerator can be, by way of example,

~~RHENOCURE RHENOCURE® AP-5, RHENOCURE RHENOCURE® AP-7, RHENOCURE RHENOCURE® AP-3, RENOCURE RHENOCURE® ZADT/G, and RHENOCURE RHENOCURE® S/G~~, which are available from Rhein Chemie Corporation of Trenton, New Jersey, and Accelerator VS, available from ~~Akro Chem~~ Akrochem Corporation. Additional cure accelerators that may be used in combination with the phosphate cure accelerator include butyl zimate; benzothiazyl disulfide (commercially known as ~~ALTEX ALTEX®~~); dithiocarbamate-containing blends (such as ~~AKROFORM AKROFORM®~~ DELTA P.M. from ~~AkroChem~~ Akrochem Corporation); and sulfides such as ~~dipentamethylenethiuram dipentamethylenethiuram~~ hexasulfide (such as ~~SULFAD SULFAD®~~ from R.T. Vanderbilt Company, Inc. of Norwalk, Connecticut). While the use of Accelerator VS was initially unacceptable in some formulations because of the foul odor problem it generated, it has also been now found that such formulations can be prepared with no significant odor if about 1.0 phr magnesium oxide is added thereto.

[0036] Suitable cure activators for the curing package include metal oxides, such as zinc oxide (e.g., TZFD-88p from Rhein Chemie Corp. ~~of Trenton, N.J.~~), magnesium oxide (e.g., ~~Elastomag~~ ELASTOMAG®-170, 170 from Morton Chemical Co.), and stearic acid (including palmitic acid), which is available from Harwick Standard Distribution Corp. of Akron, Ohio.

[0037] The carbon fibers are fibers ~~having which have been subject subjected~~ to at least partial graphitization or ~~carbonization, carbonization~~ and preferably have about 98 wt% or more carbon content. The carbon fibers should have lengths suitable for distribution in mixing equipment. Generally, the carbon fibers are preferably ~~non-continuous, noncontinuous~~ and not less than about 1/16 of an inch in length and not more than about 6 inches in length, although these ranges are not exhaustive as to the scope of the invention. Carbon fibers are supplied commercially by several companies, including ~~FORTAFIL~~ FORTAFIL® fibers (e.g., ~~FORTAFIL~~ FORTAFIL® 140 and ~~FORTAFIL~~ FORTAFIL® 144) from Akzo Nobel of Knoxville, Tennessee, carbon fibers available from Amoco of Charleston, S. ~~Carolina;~~ Carolina, and ~~PANEX~~ PANEX® 33 (1/4" x 8" or 1/4" x 15"), supplied by ~~Zoltec~~ Zoltek Corporation of St. Louis, Missouri. Generally, the carbon fibers are present in an amount of from about 2 wt% to about 50 wt%, more preferably from about 10 wt% to about 30 wt%, based on the total weight of the insulation. The amount of carbon fibers will generally vary depending on the presence of

other ingredients, such as ~~char~~char-forming agents, especially phosphate fire retardants, which supplement the carbon fibers by imparting desired physical properties to the insulation.

[0038] The carbon fibers can be used alone or in combination with other materials affecting the ablative and mechanical properties of the insulation. By way of example, suitable materials include polybenzoxazole fibers, polybenzimidazole fibers, aramid fibers, iron oxide, milled glass, silica, ceramic clay, and the like. Suitable silica particles include ~~HiSil~~ HISIL® 233 available from PPG Industries, Inc. of Lake Charles, ~~Louisiana~~ Louisiana, and ~~hydrophobized~~ hydrophobized silica particles available from Cabot Corporation of Boston, Massachusetts, as CAB-O-SIL® TS-610, CAB-O-SIL® TG-308F, CAB-O-SIL® TG-720, CAB-O-SIL® TS-500, CAB-O-SIL® TS-530, and CAB-O-SIL® TG-810G; Degussa AG of Germany as AEROSIL AEROSIL® R972, AEROSIL AEROSIL® R974, AEROSIL AEROSIL® R812, AEROSIL AEROSIL® R812S, AEROSIL AEROSIL® R711, AEROSIL AEROSIL® R 504, AEROSIL AEROSIL® R8200, AEROSIL AEROSIL® R805, AEROSIL AEROSIL® R816, AEROSIL AEROSIL® R711, and AEROSIL AEROSIL® R104; and Tulco Inc. of Ayer, Massachusetts, as TULLANOX TULLANOX® 500.

[0039] Suitable additives that may be added as required or desired include one or more of the following, in various combinations: antioxidants, flame retardants, tackifiers, plasticizers, processing aids, carbon black, pigments, and bonding agents.

[0040] Representative antioxidants for improving the longevity of the cured elastomer include, by way of example, diphenylamine reacted with acetone, available as BLE®-25 Liquid from Uniroyal Chemical Company; a mixture of mono-, di-, and tri-styrenated phenols, available as ~~AGERITE~~ AGERITE® SPAR from B.F. Goodrich Chemical Ce. Ltd. of Australia. Other suitable antioxidants include polymerized 1,2-dihydro-2,2,4-trimethylquinoline (AGERITE AGERITE® RESIN D) and mixed octylated diphenylamines (AGERITE AGERITE® STATLITES), each of which is available from R.T. Vanderbilt Co., Inc.

[0041] Fillers that function as flame retardants, or ~~char~~char-forming additives, can be used, if desired, in lesser amounts than most other additives, which makes it easier to formulate the insulation with good mechanical properties. Both inorganic and organic flame retardants are expected to be useful in the present invention. Examples of organic flame retardants include: chlorinated hydrocarbon, available as DECHLORANE®, in combination with antimony oxide

(optionally with diisodecyl phthalate (DIDP)) or hydrated alumina (such as Hydral 710 aluminum trihydrate); melamine cyanurate; phosphate and phosphate derivatives, available as PHOSCHEK P/30® PHOS-CHEK® P/30 (ammonium polyphosphate) produced by Monsanto Chemical Company of St. Louis, Missouri, which can be used alone or in combination with pentaerythritol; DECHLORANE-PLUS PLUS® 25 from Occidental Chemical Corporation of Niagara Falls, N.Y.; and silicone resin, such as DC4-7051 available through Dow Corning. An example of an inorganic flame retardant is zinc-borate, such as FIREBRAKE® ZB from U.S. Borax Inc. of Valencia, California.

[0042] Examples of suitable tackifiers are WINGTACK WINGTACK® 95 and AKROCHEM AKROCHEM® P-133. Other ingredients, such as pigments and extruder processing aids (e.g., ARMEEN ARMEEN® 18-D) well known in the art and/or suitable for use in rocket motor thermal insulation applications and extruder techniques, are intended to be included within the scope of the present invention. A suitable modifying elastomer is chlorosulfonated polyethylene, such as HYPALON 20 HYPALON®-20 available from DuPont Dow Elastomers. Non-volatile plasticizers, such as hydrocarbon oil, can also be used.

[0043] The casting of the inventive insulation into a case and curing of the inventive insulation may be performed in accordance with techniques known in the art. As referred to herein and in the appended claims, the inventive composition can be, inter alia, either applied by casting into a rocket motor case and then cured, or cured, optionally cut into appropriate geometry and size, the and then subsequently applied into the rocket motor case.

[0044] Referring now more particularly to FIGS. 2-6, the kneader in accordance with a preferred embodiment of this invention is a Buss Kneader® Buss® Kneader available through Buss Compounding Systems, AG, a plant engineering group of Georg Fischer Plant Engineering. A representative Buss Kneader® Buss® Kneader brand kneader is model MDK/E-46. This kneader is commercially available, available and is currently believed to have been used in the past in various other industries, including the following: construction; electrical and electronic component parts; automotive parts; chemicals; house appliances; foodstuffs, packaging, and consumer goods. Another similar kneader is available from B&P Process Equipment & Systems.

**[0045]** The ~~Buss-Kneader®~~ Buss® Kneader brand kneader has a housing module (or barrel) 20 defining a chamber 22. A plurality of additional modules (not shown) having respective chambers can be united together to provide an extended chamber. The housing module 20 can be equipped with a jacket or internal fluid passages for heating. In order to allow for ease in maintenance and operation, the housing module 20 can be a split-barrel arrangement to allow opening of the barrel 20 along its length, thereby facilitating access to the chamber 22.

**[0046]** In the illustrated embodiment, a single rotatable screw 24 is received in the chamber 22. Generally, the screw 24 is from about 30 mm to about 200 nm in ~~diameter~~, diameter and has a length-to-diameter (L:D) ratio of from about 8:1 to about 20:1, although this invention is not so limited, given the flexibility of uniting a desired number of modules 20.

**[0047]** As shown in FIGS. 2-6, the periphery of the screw 24 has a plurality of screw flights 30. The screw flights 30 each have a rhombic configuration in the ~~illustrate~~ illustrated embodiment, although the present invention is not thereby limited in scope. As best shown in FIG. 4, the screw flights 30 are arranged relative to each other to provide a plurality of screw flight columns 32. For each of these screw flight columns 32, the respective screw flights 30 thereof are aligned along the longitudinal axis of the screw 24, yet spaced from each other by an axial distance. In a preferred embodiment, the screw 24 has three screw flight columns 32a, 32b, and 32c. The circumferential centers C of the screw flights 30 of screw flight column 32a are positioned about the circumference at intervals of 120° from the circumferential centers of the screw flights 30 of screw flight columns 32b and 32c. Defined between each of the adjacent columns 32a, 32b, and 32c are gaps 34a, 34b, and 34c. Whereas the screw of a conventional single-screw extruder has a continuous spiral or helical screw face extending along its length, the screw 24 of the illustrated embodiment has a discontinuous screw face, with the spiral or helical path of the screw face being interpreted by the gaps 34a-c (collectively referred to as “gaps 34”).

**[0048]** The housing module 20 has kneading pins (also referred to as kneading teeth) 40, which in the illustrated embodiment have diamond-shaped cross sections. Each of the kneading pins 20 extends from an inner ~~peripheral~~ periphery thereof along a respective radial direction of the housing module 20. As shown in FIG. 4-5, the kneading pins 20 collectively define three kneading pin columns 42a, 42b, and 42c, each spaced 120° from each other about the circumference of the screw 24 and dimensioned so as to be receivable in the gaps 34. The

kneading teeth teeth 40 can be hollow and connected to a supply means for permitting the injection of fluid constituents through the kneading teeth and directly into the melt.

[0049] During operation, the screw 24 is rotated about its longitudinal axis while an axial stroke is superimposed on the screw 24 to oscillate the screw 24 back and forth in the axial direction. A gear box (not shown), also available with the ~~Buss Kneader® Buss® Kneader~~ brand kneader through Georg Fischer Plant Engineering, preferably ensures that each revolution of the screw 24 is accompanied by one full forward and backwards stroke of the screw 24. At the same time, the housing module 20 and kneading pins 40 remain stationary relative to the rotating/oscillating screw 24.

[0050] The rotating/oscillating movement of the screw 24 causes the kneading pins 40 to traverse across the faces of respective kneading screw flights 30, thus generating a shear which cleans the faces of the kneading screw flights 30 and effects dispersion and distributive mixing. This relative movement between the kneading screw flights 30 and the kneading pins 40 is explained below in more detail with reference to FIG. 6, which shows selected pins 40a and 40c and their respective paths of movement relative to the screw 24. As shown in FIG. 6, the kneading pins 40 moves across the faces of the kneading screw flights 30 and across the gaps 34, thereby cleaning the faces of the kneading screw flights 30 and causing dispersion and distributive mixing to take place.

[0051] As mentioned above, a ~~Buss Kneader® Buss® Kneader~~ model MDK/E-46 ~~brand kneader~~ having a 46 mm single screw with a process L/D ratio of 11:1 can be used. This ~~model~~ model of kneader can be used in combination with a Reliance 40 HP 1750 rpm DC Motor and Flex Pak 3000 controller.

[0052] Vertical feeds can be provided at different axial locations along the length of the housing module 20. Preferably, the inlet feeders are jacketed vertical screw feeders. Generally, the polymeric ingredients and carbon fiber are introduced into the most upstream feed, fire retardants and other additives are added further downstream (along the axial direction of the housing module 20), and the curing package is introduced at the most downstream feed port. In this manner, ~~insulation~~ the insulation composition may be continuously produced. The temperature of the chamber is generally set in the range of from about 66°C (150°F) to 93°C (200°F) during operation.

[0053] An advantage of using the kneader of this second embodiment is that the insulation composition discharged from the kneader can be introduced directly into an extruder for extrusion of the EPDM material. A suitable extruder for use with the kneader of this second embodiment is a discharge extruder GS70. The ability to extrude in this embodiment provides improvements over conventional techniques, in which the insulation composition is calendered into sheets sheets, then cut.

## EXAMPLES

[0054] The following examples illustrate embodiments which have been made in accordance with the present invention. Also set forth are comparative examples prepared for comparison purposes. The inventive embodiments are not exhaustive or exclusive, but merely representative of the many types of embodiments which may be prepared according to this invention.

[0055]

TABLE I  
(all units in parts by weight)

Ingredient	EXAMPLE						COMPARATIVE EXAMPLE	
	1	2	3	4	5	6	A	B
Trilene TRILENE® 67 [liquid EPDM]	100	100	100	100	100	100	40	
DSM-Keltan KELTAN® [solid EPDM]							50	
NORDEL NORDEL® 1040 [solid EPDM]								80
Neoprene FB [plasticizer]								20
Hypalon HYPALON® 20 [polymer]							10	
Panex PANEX® 33x8 [1/4" fibers]	26							
Fortafil FORTAFIL® 144 [carbon fibers]		38.5	45	40.5	30	25.7	26.85	
VCM [carbon fibers]								41
Akrochem AKROCHEM® P-133 [plasticizer/-tackifier/tackifier]							5	

Ingredient	EXAMPLE						COMPARATIVE EXAMPLE	
	1	2	3	4	5	6	A	B
<u>Agerite</u> AGERITE® Stalite S [antioxidant]	2	2	2	2	2	2	2	1
<u>Agerite</u> AGERITE® Resin D [antioxidant]								2
<u>HISI</u> HISIL® 233 [fire retardant filler/-char char-forming agent]	5	5	5	5	5	5	3	
<u>FireBrake</u> FIREBRAKE® ZB [fire retardant filler/-char char-forming agent]		19.5						
Hydral 710 Aluminum Trihydrate [filler]		19.5						
Carbon black [filler]								1
DC4-7051 [fire retardant/-char char-forming agent]	5			8.5	5	5	5	
Ferric oxide [filler/pigment]			1.13					
Antimony oxide (4% DIDP) [fire retardant/filler]			18					
<u>Dechlorane Plus</u> Plus® 25 [fire retardant filler/-char char-forming agent]			45					
Pentaerythritol PE 200 [fire retardant filler/-char char-forming agent]				8.5				
<u>Phos-Chek</u> PHOS-CHECK® P/40 [fire retardant filler/-char char-forming agent]				30				
Melamine cyanurate [fire retardant filler/-char char-forming agent]	25					25	25	
Zinc oxide [activator]							5	5
Kadox 920C zinc oxide [activator]	5	4	4	4	4	5		
<u>Altax</u> ALTAX® [accelerator]	1.1	1.5	1.5	1.5	1.5	1.2		

Ingredient	EXAMPLE						COMPARATIVE EXAMPLE	
	1	2	3	4	5	6	A	B
Accelerator VS	2.7							
<u>Akroform</u> <u>AKROFORM</u> ® Delta P.M. [accelerator]		0.3	0.3	0.3	0.3	0.25		
Sulfads [accelerator]		0.82	0.82	0.82	0.82	0.82		0.75
Butyl zimate [accelerator]							0.5	
<u>Rhenocure</u> <u>RHENOCUR</u> E® AP-5 [accelerator]							3.5	
<u>Crystex</u> <u>CRYSTEX®</u> OT-20 [curative]	1.05	1.22	1.22	1.22	1.22	1.22		
SP-1056 [curative]							1.1	
Captax [accelerator]								1
Tellurac [accelerator]								0.5
Sulfur [curative]								1
Total Parts by Weight	172.8	192.3	223.9	202.3	149.8	171.2	178.95	153.25

Examples 1 and 6:

[0056] All solid ingredients, with the exception of the TRILENE TRILENE® 67, were blended in a V-shell blender at ambient temperature over several hours. The TRILENE TRILENE® 67 was separately introduced into a Brabender mixer equipped with a sigma blade operating at 10 rpm and set to 60°C (140°F). The TRILENE TRILENE® 67 was mixed in the Brabender mixer for a sufficient amount of time to warm the TRILENE TRILENE® 67 to 60°C. Next, the blended material from the V-shell blender was introduced into the Brabender mixer and allowed to mix with the TRILENE TRILENE® 67 until the fibers were uniformly dispersed in the TRILENE TRILENE® 67. The formulation was then dumped from the Brabender mixer to a mill for shaping into sheets before cooling. Each sheet was about 1.27 cm (0.5 inch) in thickness.

Examples 2 through 55:

[0057] All solid ingredients, with the exception of the TRILENE TRILENE® 67 and the carbon fibers, were blended in a V-shell blender at ambient temperature over several hours. The TRILENE TRILENE® 67 was separately introduced into a Brabender mixer equipped with

a sigma blade operating at 50 rpm and set to 77°C (170°F). The ~~TRILENE~~ TRILENE® 67 was mixed in the Brabender mixer for a sufficient amount of time to warm the ~~TRILENE~~ TRILENE® 67 to 77°C. Next, the blended material from the V-shell blender was introduced into the Brabender mixer and allowed to mix with the ~~TRILENE~~ TRILENE® 67. The speed of the Brabender mixer was then slowed to 20 rpm, and the fibers were introduced into the Brabender mixer and mixed until the fibers were uniformly dispersed in the ~~TRILENE~~ TRILENE® 67. The formulation was then dumped from the Brabender mixer to a mill for shaping into sheets before cooling. Each sheet was about 1.27 cm (0.5 inch) in thickness.

Comparative Examples A and-B B:

[0058] Comparative Example A was prepared by mixing all of the ingredients, with the exception of the carbon fiber, in a laboratory mixer. The carbon fiber was incorporated into this mixture in a twin screw extruder (containing counter-rotating screws) by adding the mixed polymeric material and the carbon fibers in a single port of the twin screw extruder.

Comparative Example B was made by solvent processing with a hydrocarbon solvent.

[0059]

TABLE II

	EXAMPLE						Comparative Example	
	1	2	3	4	5	6	A	B
Average Ablation Rate for Lower Section (mm/s)	3.29	3.98	4.89	3.96	4.05	3.31	3.37	3.45
Average Ablation Rate for Middle Section (mm/s)	12.34	9.36	12.63	9.97	12.97	11.80	16.96	12.76
Average Ablation Rate for Upper Section (mm/s)	23.44	14.12	12.72	12.79	20.45	17.94	35.23	18.13

[0060] From Table II, it is seen that the inventive examples containing liquid EPDM as their exclusive EPDM ingredient (*i.e., i.e.*, no solid EPDM) exhibited comparative, comparative and and, in some instances instances, improved ablative properties to Comparative Example A

(containing less than half liquid EPDM based on the total weight of EPDM ingredients) and Comparative Example B (containing no liquid EPDM).

[0061] Examples 7 to 9 were prepared in accordance with a second embodiment of this invention by kneading the insulation in a ~~Buss Kneader®~~ Buss® Kneader. The ingredients of the insulation compositions of examples Examples 7-9 are set forth below in Table III. The ablative properties of examples Examples 7-9, a comparison of these properties to that of inventive example Example 4, are set forth below in Table IV.

[0062]

TABLE III  
(all units in parts by weight)

Ingredient	EXAMPLE		
	7	8	9
DSM-Keltan <u>KELTAN®</u> 1446A [solid EPDM]	100	100	100
Fortafil <u>FORTAFIL®</u> 243 [carbon fibers]	40.52	44.45	55.55
Cure/Filler	14.86	14.88	14.88
Fire Retardant	47.02	62.72	51.62
Total Parts by Weight	202.40	222.05	222.05

[0063]

TABLE IV

	EXAMPLE			
	7	8	9	4
Average Ablation Rate for Lower Section (mm/s)	3.68	2.85	3.44	3.35
Average Ablation Rate for Middle Section (mm/s)	9.41	8.97	9.79	9.40
Average Ablation Rate for Upper Section (mm/s)	15.42	14.75	13.61	11.64

[0064] As shown by Table IV, the insulation prepared in a ~~Buss Kneader®~~ Buss® Kneader without any liquid EPDM exhibited comparable erosion resistance to Example 4, which was prepared in a sigma mixer with liquid EPDM.

[0065] The tests were performed in a char motor, such as the one illustrated in FIG. 7. Char motors are constructed to evaluate the ablative properties of solid rocket motor case insulation materials. A char motor includes a propellant beaker 70 to provide the combustion gases, evaluation chambers to hold the test materials, and a constricting nozzle to produce the required pressure. The evaluation chamber is divided into three sections. The first one is a “low low-velocity” cylindrical region 72 about eight inches long and eight inches in diameter (approximately the same diameter as the propellant beaker). A short conical transition chamber 74 constricts the gas flow into a diameter of about 2 inches and vents the propellant gases into a 22 inch long conical test chamber. This test chamber is divided into the “mid-velocity” region 76 and “high-velocity” region 78.

[0066] Samples of insulation material to be evaluated are molded, cured, and bonded with epoxy into each of the test chambers. Prior to assembly, the cured length is determined and the thickness of each evaluation material is measured at selected intervals, nominally one inch apart. Each sample is also weighed. The samples are then assembled into the low-velocity section, the mid-velocity section, and the high-velocity section. After firing, the motor is ~~disassembled, disassembled and~~ each sample is measured again. The ablation rate is determined by subtracting the post-fired thickness of the insulation of ~~virgin insulation (i.e., i.e.,~~ after the char had been removed) at a given point from the ~~pre-prefired~~ thickness and dividing the result by the burn time of the motor. For these tests, more than one section of material was measured, and the average of all of the sections ~~are~~ is reported above.

[0067] The char motors were fired with RSRM TP-H1148 (polybutadieneacrylic acid acrylonitrile (PBAN-based)) propellant. For ~~examples~~ Examples 1-3 and comparative ~~example~~ Example A set forth in Tables I and II, the motor was fired for 12.10 seconds at an average pressure of 880 psi. For ~~examples~~ Examples 4-6 set forth in Tables I and II, the motor was fired for 11.56 seconds at an average pressure of 825 psi. Comparative ~~example~~ Example B was fired for 11.89 seconds at an average pressure of 885 psi. For ~~examples~~ Examples 7-9 and 4 set forth in Tables III and IV, the motor was fired for 12.46 seconds at an average pressure of 916.07 psi.

**[0068]** The foregoing detailed description of the invention has been provided for the purpose of explaining the principles of the invention and its practical application, thereby enabling others skilled in the art to understand the invention for various embodiments and with various modifications as are suited to the particular use contemplated. This description is not intended to be exhaustive or to limit the invention to the precise embodiments disclosed. Modifications and equivalents will be apparent to practitioners skilled in this art and are encompassed within the spirit and scope of the appended claims.

## ABSTRACT OF THE DISCLOSURE

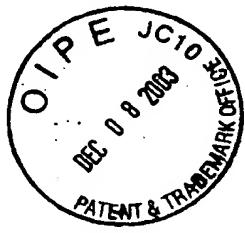
This method permits manufacturing EPDM rocket motor insulation in which carbon fibers are dispersed and immobilized in the EPDM polymeric ~~matrix~~, matrix but are not excessively fractured or fragmentized, *i.e.*, i.e., broken into smaller fragments, when encountering degrees of shear necessary to homogeneously or otherwise distribute or disperse the carbon fibers in the EPDM polymeric matrix. The method is substantially solvent ~~free~~, free and is performed via distributive/reduced shear mixing to distribute the fragile carbon fibers into a rubber matrix without excessive damage. According to one embodiment, at least about 50% of the elastomer composition introduced into the mixing apparatus is liquid EPDM terpolymer having sufficiently low molecular weight and high diene content to permit dispersion of the carbon fibers in the EPDM without substantial fragmentation of the fibers. According to another embodiment, mixing takes place in a kneader capable of rotating a screw having a discontinuous ~~screw-screw~~ thread about the screw axis while superimposing an axial axially reciprocating stroke to the screw. The kneader imparts low shear distributive mixing of the carbon fibers in the EPDM terpolymer.

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# **APPENDIX C**

**(REPLACEMENT SHEETS AND  
ANNOTATED SHEETS SHOWING CHANGES)**

**(Serial No. 09/747,192)**



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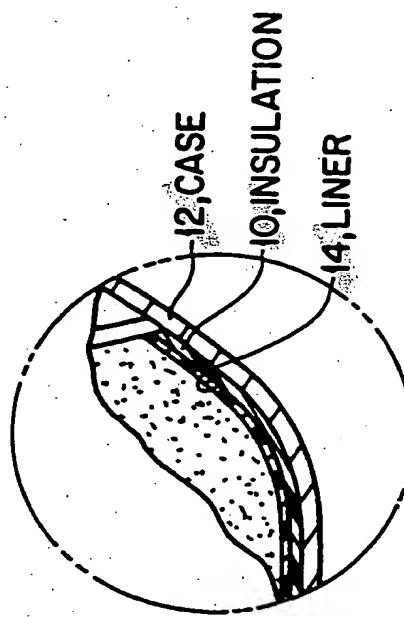


FIG. 1B

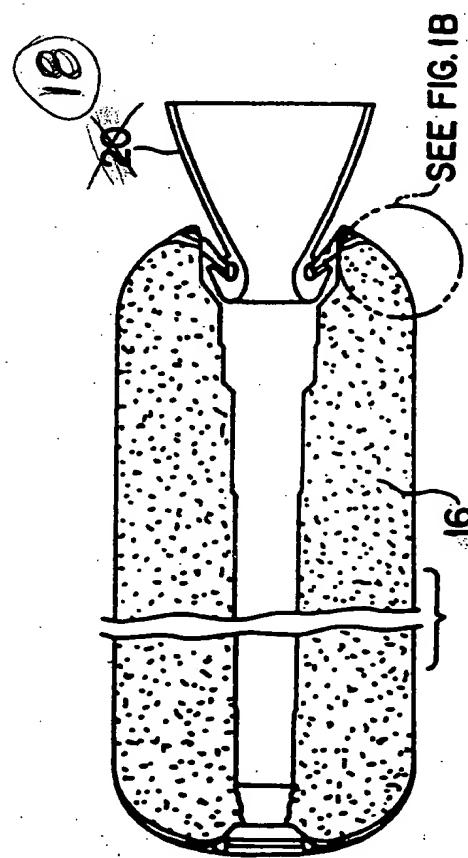
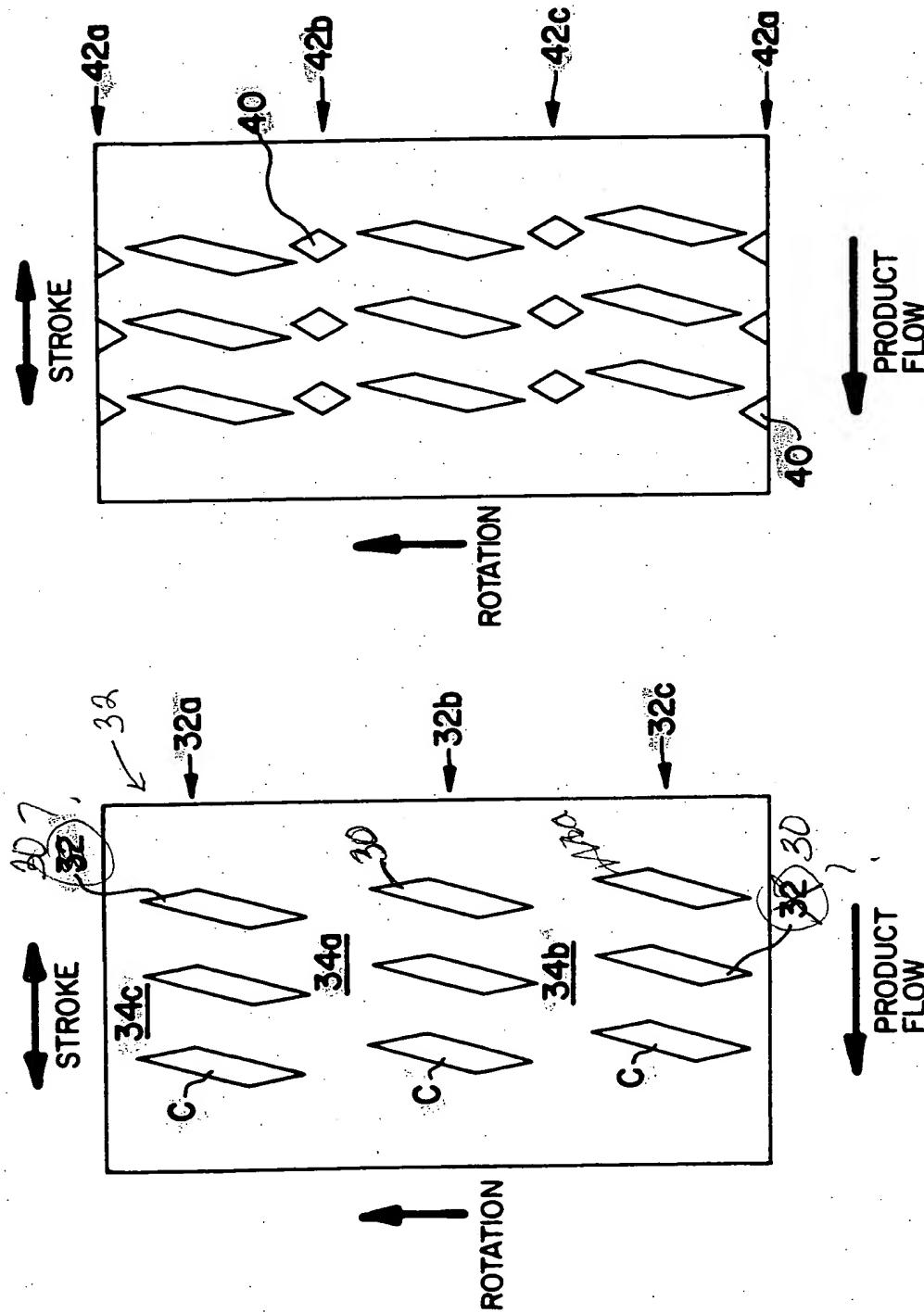


FIG. 1A



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FIG. 5FIG. 4